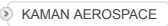
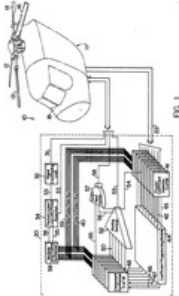
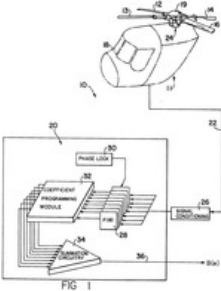
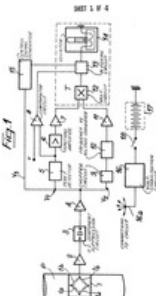
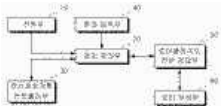
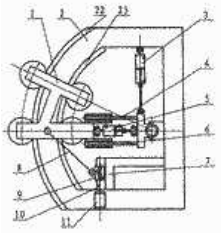
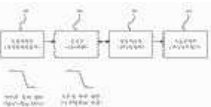
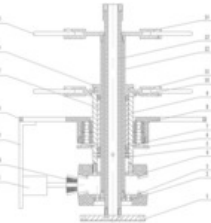
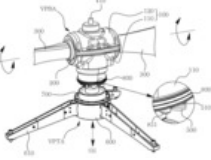
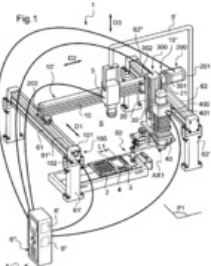
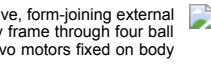
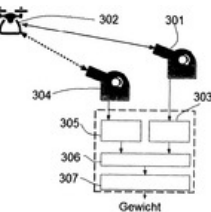


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| #                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                     |   Title | Publication number | Publ. date | Applicant/Assignee                                                                                                             |       |  |  |
|-------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------------|---------------------------------------------------------------------------------------------------------------------------------------------------------------------------|--------------------|------------|--------------------------------------------------------------------------------------------------------------------------------|-------|--|--|
| 1.                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    | <a href="#">An automatic load monitoring system with remote sensing</a>                                                                                                   | EP0339833          | 1989-11-02 |  KAMAN AEROSPACE                              | 100 % |  |  |
| <p>A novel system (10) for determining the magnitudes of loads on rotating helicopter components (12) from signals remotely measured on the helicopter fuselage (18) is characterized by a processor (20) which decomposes the measured signals into corresponding Fourier components and extracts selected components therefrom. A programming module (44) performs a multiplication on the respective Fourier components to correlate the measured components with corresponding determinative parameter values. The parameter values are established in a calibration process which is characterized by determination of the independent values of a measured signal set using a Jones orthogonalization process. The signals output from the coefficient programming module (44) are provided to a multiplying resolver (48) which receives synthesized reference sinusoids from a digital processor (38) and generates time dependent signal components therefrom. These signals are ultimately summed in an analog summer circuit (52) whose output signal corresponds to the desired parameter signal.</p>  |                                                                                                                                                                           |                    |            |                                                                                                                                |       |  |  |
| 2.                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    | <a href="#">Real time load monitoring system with remote sensing</a>                                                                                                      | EP0477948          | 1992-04-01 |  KAMAN AEROSPACE                              | 96 %  |  |  |
| <p>A system for determining the magnitude of loads on rotating helicopter components from sensor signals remotely measured on the helicopter fuselage is characterized by a controller which samples the sensor signals according to phase. These measured signals are presented to a coefficient programming module where they are combined with correlation signals to correlate the measured signal components with the desired parameter, such as flatwise bending moments of a helicopter blade. The correlation signals are established in a calibration process that determines the independent values of the measured signals using a Jones orthogonalization process. The signals output from the module are recombined to provide a time dependent signal corresponding to the desired parameter. The present system avoids decomposition and recombination of the measured signals and therefor operates in real time. &lt;IMAGE&gt;</p>                                                                                                                                                                |                                                                                                                                                                           |                    |            |                                                                                                                                |       |  |  |
| 3.                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    | <a href="#">System for determining the fatigue of a structure experiencing varying stresses</a>                                                                           | US3744300          | 1973-07-10 |  AEROSPATIALE SOCIETE NATIONALE INDUSTRIELLE | 83 %  |  |  |
| <p>A system for measuring fatigue damage in helicopter blades is based on a determination of the number of force cycles to which the blade is subjected and on the amplitude of the force experienced in each cycle.</p>                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          |                                                                                                                                                                           |                    |            |                                                                                                                                |       |  |  |
| 4.                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    | <a href="#">Checking system for rotor de-icing</a>                                                                                                                        | KR20110116773      | 2011-10-26 |  KOREA AEROSPACE INDUSTRIES                | 82 %  |  |  |
| <p>PURPOSE: An inspection system for helicopter rotor deicing is provided to inspect a real rotor deicing function regardless of inspection environment by transferring an imaginary rotor blade heating mat operating signal and an imaginary failure control signal to a robot deicing controller.<br/>CONSTITUTION: An inspection system for helicopter rotor deicing comprises an inspection control unit(20) and a rotor load unit(60). The inspection control unit inspects the operated state of a helicopter rotor deicing system on the ground by transferring an imaginary rotor blade operating signal and an imaginary failure control signal to a real rotor deicing controller(30). The rotor load unit is connected to the inspection control unit and generates the imaginary rotor blade operating signal and the imaginary failure control signal.<br/>COPYRIGHT KIPO 2012</p>                                                                                                                                                                                                                 |                                                                                                                                                                           |                    |            |                                                                                                                                |       |  |  |
| 5.                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    | <a href="#">Apparatus for fatigue test of horizontal hinge bearing of helicopter</a>                                                                                      | CN1818600          | 2006-08-16 |  UNIVERSITY YANSHAN                         | 80 %  |  |  |
| <p>A fatigue fester of horizontal hinge bearing of helicopter rotor is prepared setting form-joining an external arced form track and an internal arced form track on arc frame of support and frame short shaft on support, firm-joining shaft on frame short shaft, setting said bearing on said shaft, jacketing bearing hole of small frame on said bearing, connecting two loading hydro-cylinders separately on small frame through ball joints and connecting said hydro-cylinders with frame through a hall joint and with H form frame through ball joint.</p>                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          |                                                                                                                                                                           |                    |            |                                                                                                                                |       |  |  |

|     |                                                                                                                |                |            |                                             |      |                                                                                       |
|-----|----------------------------------------------------------------------------------------------------------------|----------------|------------|---------------------------------------------|------|---------------------------------------------------------------------------------------|
| 6.  | Monitoring system and method for loading of helicopter rotor and based on buried type optical fiber sensors    | CN109580057    | 2019-04-05 | WUHAN UNIVERSITY OF TECHNOLOGY              | 74 % |    |
| 7.  | Safety diagnosis system for a helicopter rotor using a contact-less displacement sensor                        | KR20120006807  | 2012-01-19 | KOREA AEROSPACE INDUSTRIES                  | 73 % |    |
| 8.  | Helicopter overwater forced landing model rotor lifting force simulation device                                | CN109515743    | 2019-03-26 | CHINA SPECIAL TYPE FLIER RESEARCH INSTITUTE | 70 % |    |
| 9.  | Testing apparatus for variable pitch blade assembly                                                            | KR20170024238  | 2017-03-07 | KOREA DESIGN CENTER                         | 70 % |   |
| 10. | Test bench of a device provided with a plurality of buttons and a display screen                               | EP2372335      | 2011-10-05 | EUROCOPTER                                  | 70 % |  |
| 11. | Externally swing joint-bearing fatigue experimental apparatus of automatic rotor oblique device for helicopter | CN1818598      | 2006-08-16 | UNIVERSITY YANSHAN                          | 69 % |  |
| 12. | System and methods for remote monitoring of drones and aircraft for monitoring safety and health               | DE102017108733 | 2018-09-27 | MOHAMADIN AMAL MOHAMED SAYED                | 67 % |  |